Amendments to the Claims:

Please cancel claims 1 to 7 as presented in the underlying International Application No. PCT/DE2005/000329 without prejudice.

Please add new claims as indicated in the listing of claims below.

This listing of claims will replace all prior versions, and listings, of claims in the application:

Listing of Claims:

Claims 1 to 7 (canceled).

Claim 8 (new): A rotary vane of a compressor of a gas turbine comprising:

a vane foot; and

a blade, the blade being delimited by a flow inlet edge or front edge, a flow outlet edge or rear edge, and a blade surface extending between the front edge and the rear edge and forming a suction side and a pressure side, the suction side of the blade has at least one micro-profiled or micro-structured area for optimizing flow around the blade.

Claim 9 (new): The rotary vane as recited in claim 8 wherein the at least one micro-profiled or micro-structured area is assigned to a section of the suction side of the blade, flow deceleration taking place in the section.

Claim 10 (new): The rotary vane as recited in claim 8 wherein the at least one micro-profiled or micro-structured area is assigned to a section of the suction side of the blade, the section extending over between 30% and 70% of a profile depth of the blade.

Claim 11 (new): The rotary vane as recited in claim 10 wherein the section extends over between 30% and 50% of the profile depth of the blade.

Claim 12 (new): The rotary vane as recited in claim 8 wherein the at least one micro-profiled or micro-structured area has a shark skin-like profile or structure.

Claim 13 (new): The rotary vane as recited in claim 8 wherein the vane foot includes a side having a further micro-profiled or micro-structured area structured so that the blade is strengthened in this area or that compressive stresses are induced.

Claim 14 (new): The rotary vane as recited in claim 8 wherein the vane is an aircraft engine blade.